



[4910-13-P]

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2018-0073; Product Identifier 2017-NM-100-AD]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for certain The Boeing Company Model 767-300 and -300F series airplanes. This proposed AD was prompted by reports of fatigue cracking in the lower outboard wing skin at the farthest outboard fastener of the inboard segment of a certain stringer. This proposed AD would require repetitive high frequency eddy current (HFEC) inspections for cracking of the lower outboard wing skin at the inboard segment of a certain stringer, and repair if necessary. We are proposing this AD to address the unsafe condition on these products.

DATES: We must receive comments on this proposed AD by [INSERT DATE 45 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- Federal eRulemaking Portal: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- Fax: 202-493-2251.
- Mail: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- Hand Delivery: Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Aviation Partners Boeing, 2811 S. 102nd Street, Suite 200, Seattle, WA 98168; telephone 206-762-1171; Internet <https://www.aviationpartnersboeing.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0073.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0073; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Allen Rauschendorfer, Aerospace Engineer, Airframe Section, FAA, Seattle ACO Branch, 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6450; fax: 425-917-6590; email: allen.rauschendorfer@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the ADDRESSES section. Include “Docket No. FAA-2018-0073; Product Identifier 2017-NM-100-AD” at the

beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this NPRM. We will consider all comments received by the closing date and may amend this NPRM because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this NPRM.

Discussion

We have received reports of fatigue cracking in the lower outboard wing skin at the farthest outboard fastener of the inboard segment of stringer L-9.5 on Model 767-300 airplanes with Aviation Partners Boeing winglets installed. The cracks were found at the fastener holes common to the stringer on the left- and right-hand wings. Investigation revealed that these were fatigue cracks related to Aviation Partners Boeing supplemental type certificate (STC) ST01920SE winglet retrofit kit installations. If not corrected, these cracks could extend to adjacent structure and could lead to reduced load carrying capability in the lower skin. These conditions, if not corrected, could result in failure and subsequent separation of the wing and winglet, and consequent reduced controllability of the airplane.

Related Service Information under 1 CFR part 51

We reviewed Aviation Partners Boeing (APB) Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017. The service information describes procedures for an HFEC inspection for cracking of the lower outboard wing skin at the inboard segment of stringer L-9.5, and on-condition actions that include repetitive HFEC inspections; a preventative modification (repair) that includes installing new stringers; repetitive post-modification (repair) HFEC inspections for cracking; and repair. This service information

is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

FAA's Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would require accomplishing the actions specified in the service information described previously, except as discussed under "Difference Between this Proposed Rule and the Service Information."

Difference between Proposed Rule and Service Information

Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, specifies to contact the manufacturer for instructions on how to repair certain conditions, but this proposed AD would require repairing those conditions in one of the following ways:

- In accordance with a method that we approve; or
- Using data that meet the certification basis of the airplane, and that have been approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) whom we have authorized to make those findings.

Compliance Times

The initial compliance time is the later of: 1,500 flight cycles or 7,500 flight cycles after winglet installation, whichever occurs first; or 18 months after the effective date of the AD.

The repetitive compliance times vary depending on inspection findings. The shortest repetitive interval is 1,500 flight cycles or 7,500 flight hours, whichever occurs

first. The longest repetitive interval is 6,000 flight cycles or 18,000 flight hours, whichever occurs first.

Costs of Compliance

We estimate that this proposed AD affects 140 airplanes of U.S. registry. We estimate the following costs to comply with this proposed AD:

Estimated costs – Required Actions

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
HFEC Inspections	1 work-hour X \$85 per hour = \$255, per inspection cycle	\$0	\$85, per inspection cycle	\$11,900, per inspection cycle

Estimated costs – On-Condition Actions

Action	Labor cost	Parts cost	Cost per product
Preventative Modification (Repair)	50 work-hours X \$85 per hour = \$4,250	\$0	\$4,250
Post-modification (repair) Inspections	1 work-hour X \$85 per hour = \$255	\$0	\$85

We have received no definitive data that would enable us to provide cost estimates for on-condition repairs that might be necessary as a result of the post-modification (repair) inspections specified in this proposed AD.

According to the manufacturer, some of the costs of this proposed AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all available costs in our cost estimate.

Authority for this Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

This proposed AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to transport category airplanes to the Director of the System Oversight Division.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

(1) Is not a "significant regulatory action" under Executive Order 12866,

(2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39 - AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

The Boeing Company: Docket No. FAA-2018-0073; Product Identifier 2017-NM-100-AD.

(a) Comments Due Date

We must receive comments by [INSERT DATE 45 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

(b) Affected ADs

None.

(c) Applicability

This AD applies to The Boeing Company Model 767-300 and -300F series airplanes, certificated in any category, with Aviation Partners Boeing winglets installed; as identified in Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Unsafe Condition

This AD was prompted by reports of fatigue cracking in the lower outboard wing skin at the farthest outboard fastener of the inboard segment of stringer L-9.5 on airplanes with winglets installed per Supplemental Type Certificate ST01920SE. We are issuing this AD to prevent fatigue cracking in the lower outboard wing skin, which could result in failure and subsequent separation of the wing and winglet and consequent reduced controllability of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Repetitive Inspections, Preventative Modification (Repair), Repetitive Post-Modification (Repair) Inspections, and Repair

At the applicable time specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, except as required by paragraph (h) of this AD: Do a high frequency eddy current (HFEC) inspection for cracking of the lower outboard wing skin at the inboard segment of stringer L-9.5, in accordance with Part 1 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(1) For airplanes on which “Condition 1” is found, as defined in the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin

AP767-57-013, Revision 1, dated April 11, 2017, during any inspection required by paragraph (g) or (g)(1)(i) of this AD: Do the actions required by paragraph (g)(1)(i) or (g)(1)(ii) of this AD.

(i) Repeat the inspection specified in paragraph (g) of this AD thereafter at the applicable times specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(ii) Do the actions required by paragraphs (g)(1)(ii)(A) and (g)(1)(ii)(B) of this AD:

(A) Before further flight, do the preventative modification in accordance with Part 2 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(B) At the applicable time specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, do an HFEC inspection for cracking, in accordance with Part 3 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017; and repeat the inspection thereafter at the applicable times specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(2) For airplanes on which “Condition 2” is found as defined in the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, during any inspection required by paragraph (g) or (g)(1)(i) of this AD: Do the actions required by paragraph (g)(2)(i) and (g)(2)(ii) of this AD.

(i) Before further flight, repair in accordance with Part 2 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(ii) At the applicable time specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, do an HFEC inspection for cracking, in accordance with Part 3 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017; and repeat the inspection thereafter at the applicable times specified in paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017.

(3) If any crack is found during any inspection required by paragraph (g)(1)(ii)(B) or (g)(2)(ii) of this AD, repair before further flight using a method approved in accordance with the procedures specified in paragraph (j) of this AD. Although Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, specifies to contact Boeing for repair instructions, and specifies that action as “RC” (Required for Compliance), this AD requires repair as specified in this paragraph.

(h) Exception to Service Information Specifications

Where paragraph 1.E., “Compliance,” of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, specifies a compliance time of “after the initial issue date of this service bulletin,” this AD requires compliance within the specified compliance time after the effective date of this AD.

(i) Credit for Previous Actions

For Group 2 airplanes: This paragraph provides credit for the actions specified in Part 1 and Part 2 of the Accomplishment Instructions of Aviation Partners Boeing Service Bulletin AP767-57-013, Revision 1, dated April 11, 2017, that are required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using Aviation Partners Boeing Service Bulletin AP767-57-013, dated November 30, 2016.

(j) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle ACO Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to the attention of the person identified in paragraph (k)(1) of this AD. Information may be emailed to:

9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO Branch, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) Except as required by paragraph (g)(3) of this AD: For service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (j)(4)(i) and (j)(4)(ii) of this AD apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. If a step or substep is labeled "RC Exempt," then the RC requirement is removed from that step or substep. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(k) Related Information

(1) For more information about this AD, contact Allen Rauschendorfer, Aerospace Engineer, Airframe Section, FAA, Seattle ACO Branch, 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6450; fax: 425-917-6590; email: allen.rauschendorfer@faa.gov.

(2) For service information identified in this AD, contact Aviation Partners Boeing, 2811 S. 102nd Street, Suite 200, Seattle, WA 98168; telephone 206-762-1171; Internet <https://www.aviationpartnersboeing.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on January 25, 2018.

Michael Kaszycki,
Acting Director,
System Oversight Division,
Aircraft Certification Service.
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